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Large patent application of:

John Parker

Serial No. 10/659,857

Filed September 11, 2003

VARIABLE GEOMETRY TURBINE

) Before the Examiner

) Not Yet Assigned

) Group Art Unit 3746

) Our Ref.: 27049-3

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Please find enclosed a certified copy of the United Kingdom Patent Application No. 0226943.9, which the present application claims priority to. It is believed that no fees are due for the submission of this priority document. However, if any fees are due, the Commissioner is authorized to charge Deposit Account No. 23-3030, but not to include any payment of issue fees.

Respectfully submitted,

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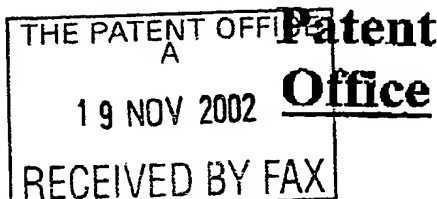
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19NOV02 E764583-1 000354
P01/7700 0.00-0226943.9**Request for grant of a patent**

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- Patents ADP number (if you know it)
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UNITED KINGDOM
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4. Title of the invention
VARIABLE GEOMETRY TURBINE
-
5. Name of your agent (if you have one) Marks & Clerk

 "Address for service" in the United Kingdom to which all correspondence should be sent (including the postcode) Sussex House
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9 /

Claim(s)

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Statement of Inventorship and right to grant of a patent (Patents Form 7/77)

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11.

I/We request the grant of a patent on the basis of this application.

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DUPLICATE

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VARIABLE GEOMETRY TURBINE

The present invention relates to a variable geometry turbine. Particularly, but not exclusively, the invention relates to the turbine of a turbocharger for an internal combustion engine. More particularly still, the invention relates to vehicle engine turbochargers which may be controlled to operate as an engine exhaust brake.

Turbochargers are well known devices for supplying air to the intake of an internal combustion engine at pressures above atmospheric (boost pressures). A conventional turbocharger essentially comprises an exhaust gas driven turbine wheel mounted on a rotatable shaft within a turbine housing. Rotation of the turbine wheel rotates a compressor wheel mounted on the other end of the shaft within a compressor housing. The compressor wheel delivers compressed air to the engine intake manifold. The turbocharger shaft is conventionally supported by journal and thrust bearings, including appropriate lubricating systems, located within a central bearing housing connected between the turbine and compressor wheel housings.

In known turbochargers, the turbine stage comprises a turbine chamber within which the turbine wheel is mounted, an annular inlet passageway defined between facing radial walls arranged around the turbine chamber, an inlet arranged around the inlet passageway, and an outlet passageway extending from the turbine chamber. The passageways and chambers communicate such that pressurised exhaust gas admitted to the inlet chamber flows through the inlet passageway to the outlet passageway via the turbine and rotates the turbine wheel. It is also well known to trim turbine performance by providing vanes, referred to as nozzle vanes, in the inlet passageway so as to deflect gas flowing through the inlet passageway towards the direction of rotation of the turbine wheel.

Turbines may be of a fixed or variable geometry type. Variable geometry turbines differ from fixed geometry turbines in that the size of the inlet passageway can be varied to optimise gas flow velocities over a range of mass flow rates so that the power output of the turbine can be varied to suite varying engine demands. For instance, when the volume of exhaust gas being delivered to the turbine is relatively

low, the velocity of the gas reaching the turbine wheel is maintained at a level which ensures efficient turbine operation by reducing the size of the annular inlet passageway.

In one known type of variable geometry turbine, one wall of the inlet passageway is defined by an axially moveable wall member, generally referred to as a nozzle ring. The position of the nozzle ring relative to a facing wall of the inlet passageway is adjustable to control the axial width of the inlet passageway. Thus, for example, as gas flowing through the turbine decreases the inlet passageway width may also be decreased to maintain gas velocity and optimise turbine output. Such nozzle rings essentially comprise a radially extending wall and inner and outer axially extending annular flanges. The annular flanges extend into an annular cavity defined in the turbine housing (a part of the housing which in practice be provided by the bearing housing) which accommodates axial movement of the nozzle ring.

The nozzle ring may be provided with vanes which extend into the inlet passageway and through slots provided on the facing wall of the inlet passageway to accommodate movement of the nozzle ring. Alternatively, vanes may extend from the fixed wall through slots provided in the nozzle ring. Generally the nozzle ring is supported on rods extending parallel to the axis of rotation of the turbine wheel and is moved by an actuator which axially displaces the rods. Various forms of actuator are known for use in variable geometry turbines, including pneumatic, hydraulic and electric actuators, mounted externally of the turbocharger and connected to the variable geometry system via appropriate linkages.

In addition to the conventional control of a variable geometry turbine to optimise turbocharger performance, it is also known to take advantage of the facility to minimise the turbocharger inlet to provide an exhaust braking function. Exhaust brake systems of various forms are widely fitted to vehicle engine systems, in particular to compression ignition engines (diesel engines) used to power large vehicles such as trucks. Conventional exhaust brake systems comprise a valve in the exhaust line from the engine which when activated substantially blocks the engine exhaust (fully blocking the exhaust line would stall the engine). This creates back

pressure which retards rotation of the engine providing a braking force which is transmitted to the vehicle wheels through the vehicle drive train. The exhaust braking may be employed to enhance the effect of the conventional friction brakes acting on the vehicle wheels, or in some circumstances be used independently of the normal wheel braking system, for instance to control down hill speed of a vehicle. With some exhaust brake systems the brake is set to activate automatically when the engine throttle is closed (i.e. when the driver lifts his foot from the throttle pedal), and in others the exhaust brake may require manual activation by the driver, such as depression of a separate brake pedal. The exhaust brake valve is generally controllable to modulate the braking effect, for example to maintain a constant vehicle speed.

With a variable geometry turbine it is not necessary to provide a separate exhaust brake valve. Rather, the turbine inlet passageway may simply be closed to its minimum flow area when braking is required. The level of braking may be modulated by control of the inlet passageway size by appropriate control of the axial position of the nozzle ring (or other variable geometry mechanism). Whilst having the advantage of obviating the need to provide a separate exhaust brake valve, there are however problems associated with operation of variable geometry turbines in an exhaust braking mode.

In particular with the modern highly efficient turbines, a relatively high air flow is still delivered to the engine as the inlet passageway is reduced towards the minimum width. This can result in engine cylinder pressures approaching or exceeding acceptable limits if the inlet passage is closed too far. Accordingly there is a practical limit on the extent to which the inlet passage can be closed in braking mode, which in turn limits the effective braking force that can be provided by control of a conventional variable geometry turbine.

It is an object of the present invention to obviate or mitigate the above disadvantage.

According to the present invention there is provided a variable geometry turbine comprising a turbine wheel supported in a housing for rotation about a turbine

axis, an annular inlet passage way extending radially inwards towards the turbine wheel, the annular inlet passageway being defined between a radial wall of a moveable wall member and a facing wall of the housing, the moveable wall member being mounted within an annular cavity provided within the housing and having inner and outer annular surfaces, the wall member being moveable axially between first and second positions to vary the width of the inlet passage way, the second axial position being closer to the said facing wall of the housing than the first axial position, the moveable wall member having a first annular flange extending axially from the radial wall into said cavity in a direction away from said facing wall of the housing, a first annular seal being disposed between said first annular flange and the adjacent inner or outer annular surface of the cavity, said first annular seal being mounted to one of said first annular flange or said adjacent annular surface of the cavity;

wherein one or more inlet bypass passages are provided in the other of said first annular flange and said adjacent cavity surface, such that said first annular seal and bypass passageways move axially relative to one another as the moveable wall member moves between said first and second positions; and

wherein said first annular seal and the or each bypass passage are axially located such that with the annular wall member in said first position the seal prevents exhaust gas flow through the cavity but with said moveable wall member in the second position the or each bypass passage permits the flow of exhaust gas through said cavity to the turbine wheel thereby bypassing the annular inlet passageway.

Embodiments of the present invention will now be described, by way of example only, with reference to the accompanying drawings, in which:

Figure 1 is a cross-sectional illustration of a prior art turbocharger;

Figures 2a and 2b illustrate a modification of the turbocharger of Figure 1 in accordance with the present invention;

Figures 3a and 3b illustrate a second embodiment of the present invention; and

Figures 4a and 4b schematically illustrate a third embodiment of the present invention.

Referring to Figure 1, this illustrates a known turbocharger as disclosed in US patent number 5,044,880. The turbocharger comprises a turbine stage 1 and a compressor stage 2. The turbine stage 1 is a variable geometry turbine comprising a turbine housing 3 defining a volute or inlet chamber 4 to which exhaust gas from an internal combustion engine (not shown) is delivered. The exhaust gas flows from the inlet chamber 4 to an outlet passageway 5 via an annular inlet passageway 6 defined on one side by a radial wall 7 of a moveable annular member 8, referred to herein as a nozzle ring, and on the other side by a facing radial wall 9 of the housing 3. An array of nozzle vanes 10 extend through slots in the nozzle ring 8 across the inlet passageway 6 from a vane support ring 11 which is mounted on support pins 12. The arrangement is such that the degree to which the vanes 10 extend across the inlet passageway 6 is controllable independently of the nozzle ring 8 and will not be described in detail here.

Gas flowing from the inlet chamber 4 to the outlet passageway 5 passes over a turbine wheel 12 which as a result drives a compressor wheel 13 via turbocharger shaft 14 which rotates on bearing assemblies 15 located within a bearing housing 16 which connects the turbine housing 2 to a compressor housing 17. Rotation of the compressor wheel 13 draws in air through a compressor inlet 18, and delivers compressed air to the intake of the engine (not shown) via an outlet volute 19. It will be appreciated that the bearing housing also houses oil supply and seal arrangements, the details of which are not necessary for an understanding of the present invention.

The nozzle ring 8 comprises a radially extending annular portion defining the radial wall 7, and axially extending inner and outer annular flanges 20 and 21 respectively which extend into an annular cavity 22 provided in the turbine housing 3. With the turbine construction shown in the figures, the majority of the cavity 22 is in fact defined by the bearing housing 16 – this is purely a result of the construction of the particular turbocharger to which the invention is in this instance is applied and for the purposes of the present invention no distinction is made between the turbine housing and bearing housing in this regard. -The cavity 22 has a radially extending annular opening 23 defined between radially inner and outer annular surfaces 24 and

25. A seal ring 26 is located in an annular groove provided in outer annular surface 25 and bears against the outer annular flange 21 of the nozzle ring 8 to prevent exhaust gas flowing through the turbine via the cavity 22 rather than the inlet passageway 6.

A pneumatically operated actuator 27 is operable to control the position of the nozzle ring 8 via an actuator output shaft 28 which is linked to a stirrup member 29 which in turn engages axially extending guide rods 30 (only one of which is visible in the figures) which support the nozzle ring 8 via linking plates 31. Accordingly, by appropriate control of the actuator 27 the axial position of the guide rods and thus of the nozzle ring 8 can be controlled. Figure 1 shows the nozzle ring 8 in its fully open position in which the inlet passageway 6 is at its maximum width.

As mentioned above, a variable geometry turbine such as that disclosed in Figure 1 can be operated to function as an exhaust brake by closing the inlet passageway 6 to a minimum width when braking force is required. However, also as mentioned above, with such an arrangement the minimum width of the inlet passageway under exhaust braking conditions is limited by the need to avoid unacceptably high engine cylinder pressures.

Figures 2a and 2b illustrate a modification of the turbocharger of Figure 1 in accordance with the present invention. Only those parts of the turbine which need to be described for an understanding of the invention are shown in Figures 2a and 2b which are enlargements of the nozzle ring/inlet passageway region of the turbocharger showing the nozzle ring in fully open and fully closed positions respectively. The nozzle ring 8 is modified by the provision of a circumferential array of apertures 32 provided through the radially outer flange 21. The positioning of the apertures 32 is such that they lie on the side of the seal ring 26 remote from the inlet passageway 6 (as shown in Figure 2a) except when the nozzle ring 8 approaches the closed position, at which point the apertures 32 pass the seal 26 (as shown in Figure 2b). This opens bypass flow path allowing some exhaust gas to flow from the inlet chamber 4 to the turbine wheel 12 via the cavity 22 rather than through the inlet passageway 6. The exhaust gas flow that bypasses the inlet passageway 6, and nozzle vanes 10, will do

less work than the exhaust gas flow through the inlet passageway 6 particularly since this is turned in a tangential direction by the vanes 10. In other words, as soon as the apertures 32 are brought into communication with the inlet passageway 6 there is an immediate reduction in the efficiency of the turbocharger and corresponding drop in compressor outflow pressure (boost pressure) with an accompanying drop in engine cylinder pressure.

Thus, with the present invention the provision of the inlet bypass apertures 32 will have no effect on the efficiency of the turbocharger under normal operating conditions but when the turbine is operated in an engine braking mode, and the inlet passageway is reduced to its minimum, the apertures will 32 facilitate a greater reduction in inlet passageway size than is possible with the prior art without over pressurising the engine cylinders. This thereby provides improved engine braking performance.

It will be appreciated that the efficiency reducing effect on the turbocharger can be predetermined by appropriate selection of the number, size, shape and position of the apertures 32.

Figures 3a and 3b illustrate a second embodiment of the present invention. As with Figures 2a and 2b, only detail of the nozzle ring/inlet passageway region of the turbine is illustrated. Where appropriate, the same reference numerals are used in Figures 3a and 3b as used in Figures 1 and 2. Figures 3a and 3b illustrate application of the invention to an otherwise conventional turbine which differs from the turbine of Figure 1 in several respects. Firstly, the nozzle vanes 10 are mounted on the nozzle ring 8 and extend across the inlet passageway 6 and into a cavity 33 via respective slots provided in a shroud plate 34 which together with the radial wall 7 of the nozzle ring 8 defines the width of the inlet passageway 6. This is a well known arrangement.

Secondly, in accordance with the teaching of European patent number 0 654 587, pressure balancing apertures 35 are provided through the radial wall 7 of the nozzle ring 8 and the inner annular flange 20 is sealed with respect to the housing 3 by a respective seal ring 36 located in an annular groove provided in the radially inner annular portion 24 of the housing 3. The provision of the apertures 35 ensures that

pressure within the cavity 22 is equal to the static pressure applied to the radial face 7 of the nozzle ring 8 by exhaust gas flow through the inlet passageway 6. This reduces the load on the nozzle ring with an increase in the accuracy of control of the position of the nozzle ring 8, particularly as the inlet passageway 6 is reduced towards its minimum width.

In view of the provision of a radially inner seal ring 36, application of the present invention requires provision of gas bypass passages 32a and 32b in both the outer and inner annular flanges 20 and 21 of the nozzle ring 8. The passages 32a and 32b are positioned relative to the respective seal rings 26 and 36 so that they open into communication with the inlet passageway side of the seal rings 23, 36 at the same time thereby providing a bypass flow passage through the cavity 22 achieving exactly the same effect as described above in relation to the embodiment of Figures 2a and 2b.

It is also known to seal the nozzle ring with respect of the housing by locating inner and/or outer seal rings within locating grooves provided on the nozzle ring rather than locating grooves provided within the housing. In this case the seal ring(s) will move with the nozzle ring. Specifically, Figures 4a and 4b illustrate the nozzle ring/inlet passageway region of the turbine disclosed in European patent number 0 654 587 (mentioned above) modified in accordance with the present invention. Where appropriate, the same reference numerals are used in Figures 4a and 4b as are used above. As with the turbine arrangement of Figures 3a and 3b, the nozzle vanes 10 are supported by the nozzle ring 8 and extend across the inlet passageway 6, through a shroud plate 34 and into a cavity 33. Pressure balancing apertures 35 are provided through the radial wall 7 of the nozzle ring 8, which is sealed with respect to the cavity 22 by inner and outer ring seals 26 and 37. However, whereas the seal ring 26 is located within a groove provided in the housing 3, the radially outer seal ring 37 is located within a groove 38 provided within the outer annular flange 21 of the nozzle ring 8 and thus moves as the nozzle ring moves.

In accordance with the present invention the inner annular flange 20 of the nozzle ring 8 is provided with inlet bypass apertures 32 which pass the seal ring 26 as the nozzle ring moves to close the inlet passageway 6 to a minimum (as illustrated in Figure 4b).

However, the outer inlet bypass path is provided not by apertures through the nozzle ring, but by a circumferential array of recesses 3a formed in the outer annular portion 25 of the opening 23 of cavity 22. As can be seen from Figure 4a, under normal operating conditions the seal ring 37 will be disposed inward of the recesses 39 preventing the passage of exhaust gas around the nozzle ring 8 and through the cavity 22. However, as the nozzle ring moves to close the inlet passageway 6 to a minimum, as shown in Figure 4b, the seal ring 37 moves into axial alignment with the recesses 39 which thereby provide a bypass path around the seal ring 37 to allow gas to flow through the cavity 22, and to the turbine wheel via the inlet bypass apertures 32 provided in the inner annular flange of the nozzle ring 8. It will be appreciated that the effect of the recesses 38 is directly equivalent to the effect of apertures 32 and that in operation this embodiment of the invention will function in substantially the same way as the other embodiments of the invention described above.

It will be appreciated that modifications may be made to the embodiments of the invention described above. For instance, if only one seal ring is required as for example in embodiment of Figures 8, and this is located on the nozzle ring, then there will be no need to provide aperture 32 in the inner flange of the nozzle ring. Similarly, if there are both inner and outer seal rings located in the housing, it will be necessary to provide bypass recesses in both the inner and outer annular portions of the housing instead of bypass apertures through the nozzle ring.

Other possible modifications and applications of the present invention will be readily apparent to the appropriately skilled person.

CLAIMS

1. A variable geometry turbine comprising a turbine wheel supported in a housing for rotation about a turbine axis, an annular inlet passage way extending radially inwards towards the turbine wheel, the annular inlet passageway being defined between a radial wall of a moveable wall member and a facing wall of the housing, the moveable wall member being mounted within an annular cavity provided within the housing and having inner and outer annular surfaces, the wall member being moveable axially between first and second positions to vary the width of the inlet passage way, the second axial position being closer to the said facing wall of the housing than the first axial position, the moveable wall member having a first annular flange extending axially from the radial wall into said cavity in a direction away from said facing wall of the housing, a first annular seal being disposed between said first annular flange and the adjacent inner or outer annular surface of the cavity, said first annular seal being mounted to one of said first annular flange or said adjacent annular surface of the cavity;

wherein one or more inlet bypass passages are provided in the other of said first annular flange and said adjacent cavity surface, such that said first annular seal and bypass passageways move axially relative to one another as the moveable wall member moves between said first and second positions; and

wherein said first annular seal and the or each bypass passage are axially located such that with the annular wall member in said first position the seal prevents exhaust gas flow through the cavity but with said moveable wall member in the second position the or each bypass passage permits the flow of exhaust gas through said cavity to the turbine wheel thereby bypassing the annular inlet passageway.

2. A variable geometry turbine according to claim 1, wherein the annular seal is mounted to said annular surface of the cavity, and the or each bypass passage comprises an aperture through said first annular flange of the moveable wall member.

3. A variable geometry turbine according to claim 2, wherein the annular seal is located within an annular groove provided within said annular surface of the cavity.
4. A variable geometry turbine according to claim 1, wherein said annular seal is mounted to said first annular flange, and the or each bypass passage comprises a recess provided in the adjacent annular surface of the cavity.
5. A variable geometry turbine according to claim 4, wherein said annular seal is located within an annular groove provided in an outer surface of said first annular flange.
6. A variable geometry turbine according to any preceding claim, wherein said first annular flange extends axially from the radially outermost periphery of the radial wall of the moveable wall member, and said first annular seal is disposed between the first annular flange and said outer surface of the cavity.
7. A variable geometry turbine according to claim 6, wherein said moveable wall member comprises a second annular flange extending axially into said cavity from the radially innermost periphery of the radial wall, and a second annular seal is mounted to one of said second annular flange or said inner annular surface of the cavity, and one or more inlet bypass passages are provided in the other of said second annular flange and said inner cavity surface.
8. A variable geometry turbine according to claim 7, wherein apertures are provided through said radial wall of the moveable wall member whereby the cavity is in fluid communication with the inlet passageway, said apertures being arranged such that in use a resultant force is exerted on the moveable wall member which force is always in a single axial direction.

9. A variable geometry turbine according to any preceding claim, wherein when the moveable wall member is in said second position the width of the inlet passageway is at a minimum.
10. A variable geometry turbine according to any preceding claim, wherein when the moveable wall member is between said first position and a third position intermediate the first and second positions the or each seal prevents flow of gas through said cavity via said bypass passages, and wherein when said moveable wall member is between said third position and said second position said bypass passages permit the flow of gas through said cavity, and wherein said third position is closer to said second position than to said first position.
11. A variable geometry turbine according to claim 10, wherein the positions between said first and third positions correspond to a normal high efficiency operating mode of the turbine, and positions between said third and second position correspond to an exhaust braking mode of operation of the turbine.
12. A variable geometry turbine according to any preceding claim, comprising a plurality of said bypass passages circumferentially arranged around the respective annular flange or annular cavity surface.
13. A variable geometry turbine according to any preceding claim, comprising nozzle vanes extending into the inlet passage way.
14. A variable geometry turbine according to any preceding claim, wherein said nozzle vanes extend from said nozzle ring.
15. A turbocharger comprising a variable geometry turbine according to any preceding claim.

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16. A turbocharged internal combustion engine comprising a turbocharger according to claim 13.

17. A powered vehicle comprising a turbocharged internal combustion engine according to claim 14, wherein operating said turbocharger with the annular wall member at or adjacent said second position provides an exhaust braking function.

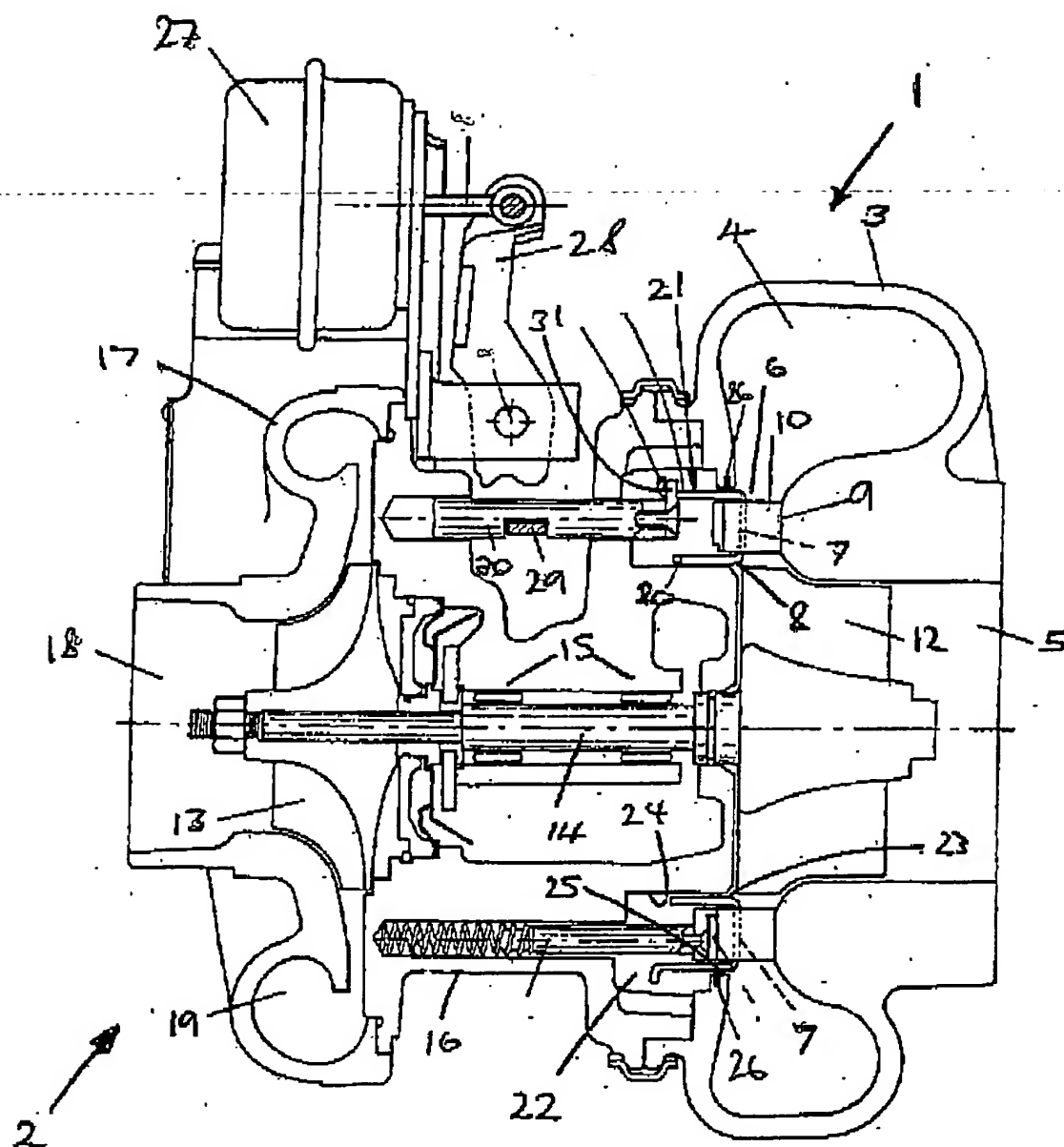


Fig 1



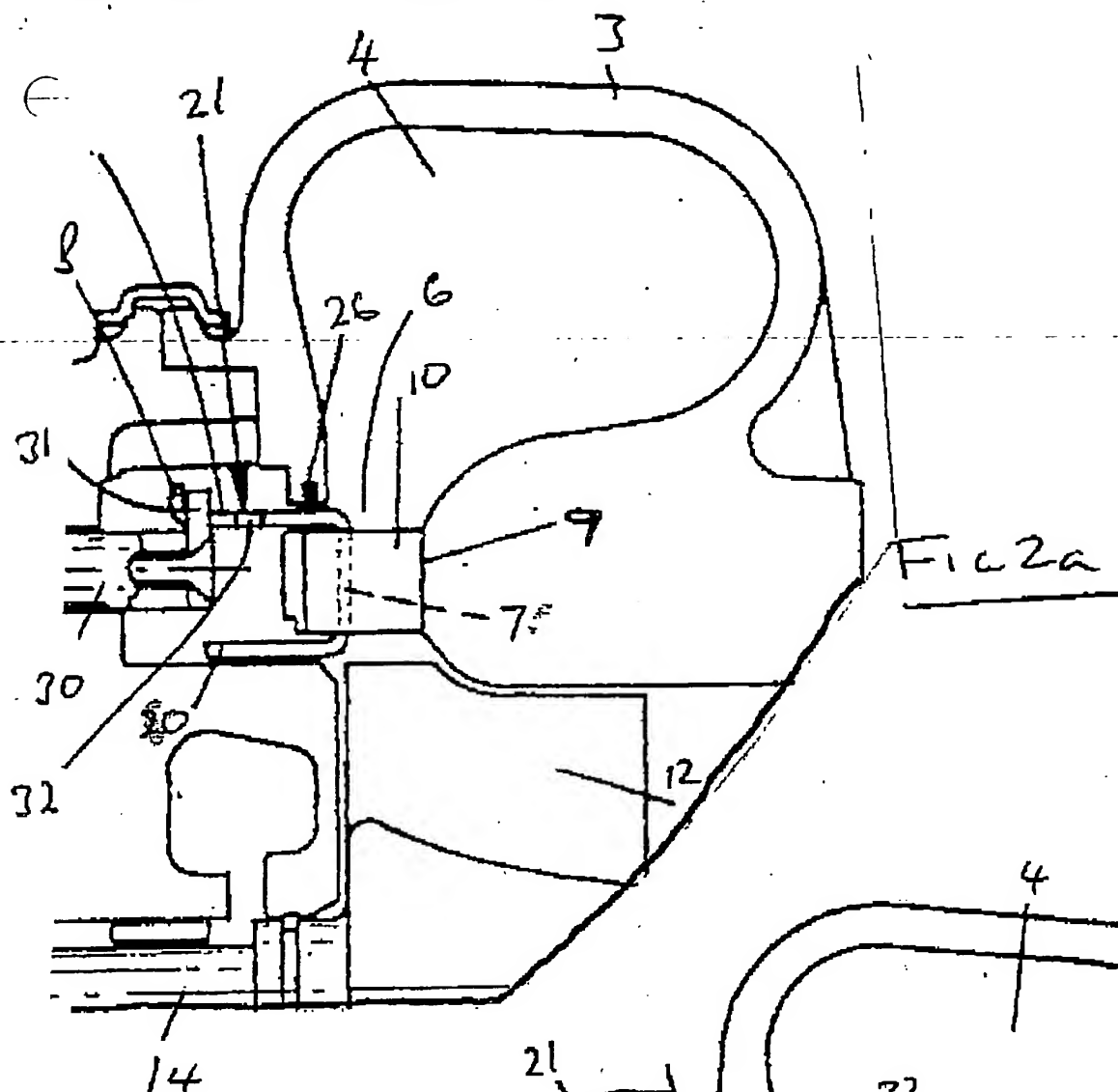
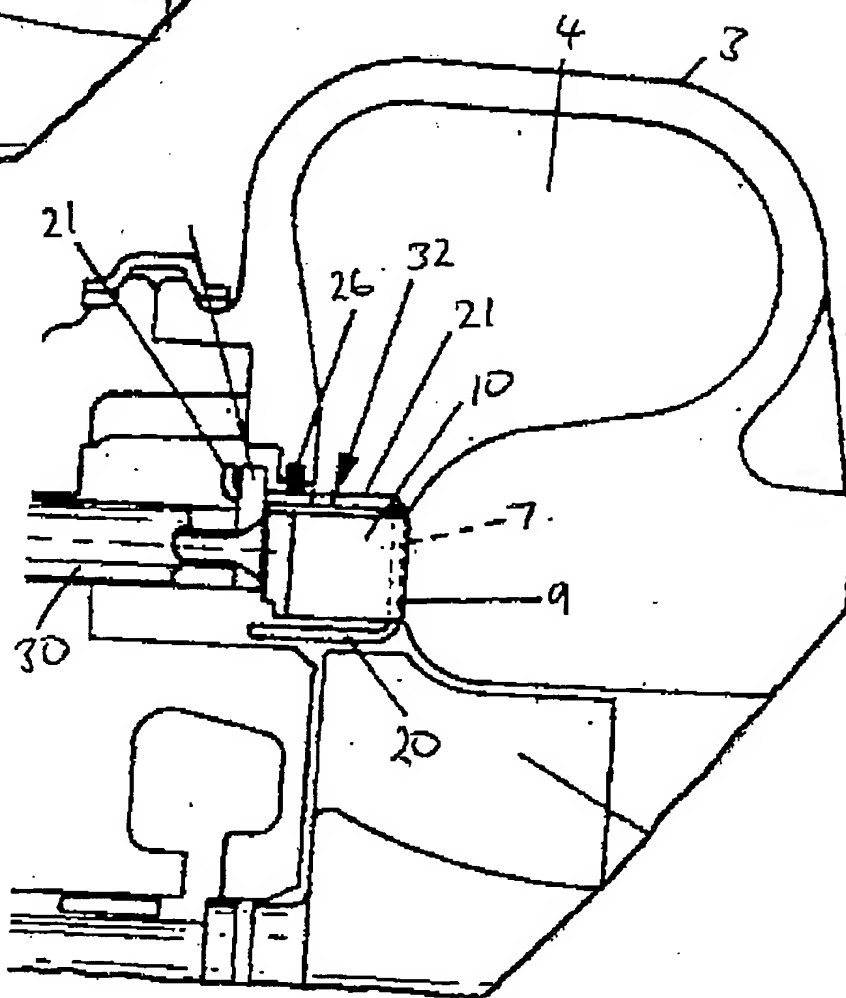
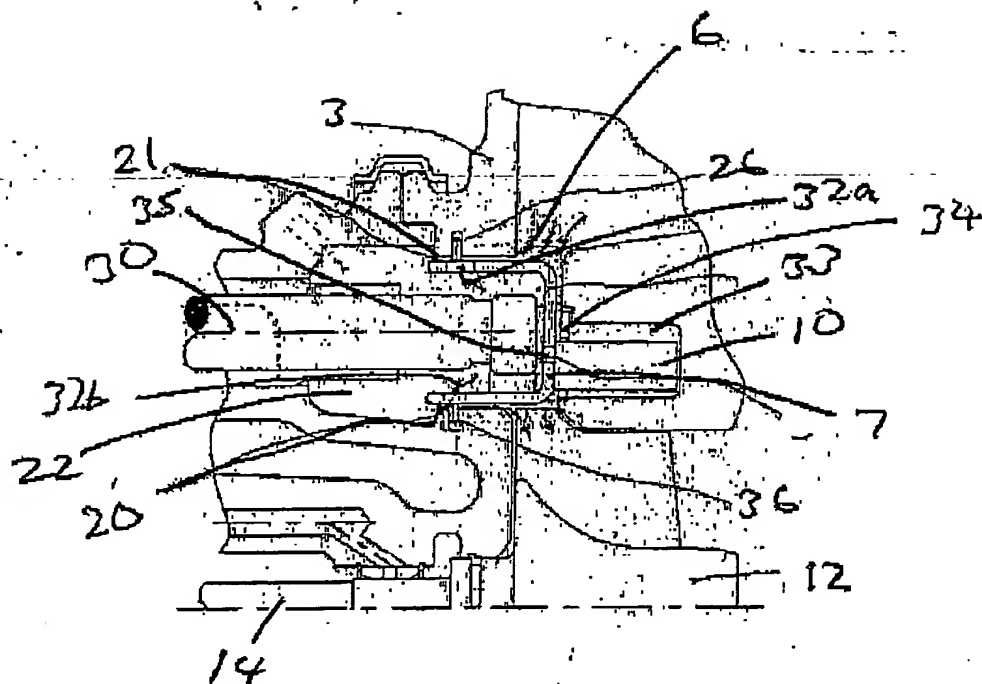
Fig 2b

Fig 7aFig 7b